

ABSTRACT OF THE DISCLOSURE

In the blade structure in a gas turbine, front-edge
including angles are made large. As a result, a curve of
a relative relationship between incidence angles i_{c1} and
5 i_{s1} and pressure loss becomes mild. Entrance metal angles
are made small. As a result, it becomes possible to make
the incidence angles small. Chord length of a chip portion
of a moving blade is made large. As a result, it becomes
possible to make small the deceleration on a rear surface
10 of the chip portion of the moving blade. Accordingly, it
becomes possible to make the pressure loss small, and
therefore, it becomes possible to improve the turbine
efficiency.

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